

EE 570: Location and Navigation

Aided INS

Aly El-Osery Kevin Wedeward

Electrical Engineering Department, New Mexico Tech
Socorro, New Mexico, USA

In Collaboration with
Stephen Bruder
Electrical and Computer Engineering Department
Embry-Riddle Aeronautical University
Prescott, Arizona, USA

April 13, 2016

- Truth value

 \vec{x}

- Measured value

 $\tilde{\vec{x}}$

- Estimated or computed value

 $\hat{\vec{x}}$

- Error

$$\delta\vec{x} = \vec{x} - \hat{\vec{x}}$$

Initially the accelerometer and gyroscope measurements, $\tilde{\vec{f}}_{ib}^b$ and $\tilde{\vec{\omega}}_{ib}^b$, respectively, will be modeled as

$$\tilde{\vec{f}}_{ib}^b = \vec{f}_{ib}^b + \Delta\vec{f}_{ib}^b = \hat{\vec{f}}_{ib}^b + \Delta\hat{\vec{f}}_{ib}^b \quad (1)$$

$$\tilde{\vec{\omega}}_{ib}^b = \vec{\omega}_{ib}^b + \Delta\vec{\omega}_{ib}^b = \hat{\vec{\omega}}_{ib}^b + \Delta\hat{\vec{\omega}}_{ib}^b \quad (2)$$

where \vec{f}_{ib}^b and $\vec{\omega}_{ib}^b$ are the specific force and angular rates, respectively; and $\Delta\vec{f}_{ib}^b$ and $\Delta\vec{\omega}_{ib}^b$ represents the errors. In later lectures we will discuss more detailed description of these errors.

Accelerometers

$$\tilde{\vec{f}}_{ib}^b = \vec{b}_a + (\mathcal{I} + M_a)\vec{f}_{ib}^b + \vec{n}l_a + \vec{w}_a \quad (3)$$

Gyroscopes

$$\tilde{\vec{\omega}}_{ib}^b = \vec{b}_g + (\mathcal{I} + M_g)\vec{\omega}_{ib}^b + G_g\vec{f}_{ib}^b + \vec{w}_g \quad (4)$$

- Position error

$$\delta \vec{r}_{\beta b}^{\gamma} = \vec{r}_{\beta b}^{\gamma} - \hat{\vec{r}}_{\beta b}^{\gamma} \quad (5)$$

- Velocity error

$$\delta \vec{v}_{\beta b}^{\gamma} = \vec{v}_{\beta b}^{\gamma} - \hat{\vec{v}}_{\beta b}^{\gamma} \quad (6)$$

- Specific force errors

$$\delta \vec{f}_{ib}^b = \vec{f}_{ib}^b - \hat{\vec{f}}_{ib}^b \quad (7)$$

$$\Delta_e \vec{f}_{ib}^b = \Delta \vec{f}_{ib}^b - \Delta \hat{\vec{f}}_{ib}^b = -\delta \vec{f}_{ib}^b \quad (8)$$

- Angular rate errors

$$\delta \vec{\omega}_{ib}^b = \vec{\omega}_{ib}^b - \hat{\vec{\omega}}_{ib}^b \quad (9)$$

$$\Delta_e \vec{\omega}_{ib}^b = \Delta \vec{\omega}_{ib}^b - \Delta \hat{\vec{\omega}}_{ib}^b = -\delta \vec{\omega}_{ib}^b \quad (10)$$

Recall

$$\begin{pmatrix} \delta \dot{\psi}_{eb}^e \\ \delta \dot{\mathbf{v}}_{eb}^e \\ \delta \dot{\mathbf{r}}_{eb}^e \end{pmatrix} = \begin{bmatrix} -\Omega_{ie}^e & 0_{3 \times 3} & 0_{3 \times 3} \\ -[\hat{\mathbf{C}}_b^e \hat{\mathbf{f}}_{ib}^b \times] & -2\Omega_{ie}^e & \frac{2g_0(\hat{L}_b)}{r_{eS}^e(\hat{L}_b)} \frac{\hat{\mathbf{r}}_{eb}^e}{|\hat{\mathbf{r}}_{eb}^e|^2} (\hat{\mathbf{r}}_{eb}^e)^T \\ 0_{3 \times 3} & \mathcal{I}_{3 \times 3} & 0_{3 \times 3} \end{bmatrix} \begin{pmatrix} \delta \psi_{eb}^e \\ \delta \mathbf{v}_{eb}^e \\ \delta \mathbf{r}_{eb}^e \end{pmatrix} + \begin{bmatrix} 0 & -\hat{\mathbf{C}}_b^e \\ -\hat{\mathbf{C}}_b^e & 0 \\ 0 & 0 \end{bmatrix} \begin{pmatrix} \Delta_e \vec{f}_{ib}^b \\ \Delta_e \vec{\omega}_{ib}^b \end{pmatrix} \quad (11)$$

In reality there will be error terms in the sensor that can not be calibrated. These terms may be estimated. The error in the estimation of these terms may be expressed as

$$\Delta_e \vec{f}_{ib}^b = \Delta \vec{f}_{ib}^b - \Delta \hat{\vec{f}}_{ib}^b = F_{va} \delta \vec{X}_a + \vec{c}_a \quad (12)$$

$$\Delta_e \vec{\omega}_{ib}^b = \Delta \vec{\omega}_{ib}^b - \Delta \hat{\vec{\omega}}_{ib}^b = F_{\psi g} \delta \vec{X}_g + \vec{c}_g \quad (13)$$

These terms represent the difference between what we estimate the errors in the sensors to be (either through calibration or online estimation) and the actual errors in the sensor.

The matrices F_{va} and $F_{\psi g}$, depend on the needed level of complexity in modeling the errors. For example if we only model biases, e.g., $\delta\vec{x}_a = \delta\vec{b}_a$, then $F_{va} = \mathcal{I}$.

If more error terms are modeled, then most likely, we will end up with non-linear equations, and therefore linearization is necessary.

$$\delta \dot{\vec{x}}_a = F_{aa} \delta \vec{x}_a + \vec{w}_a \quad (14)$$

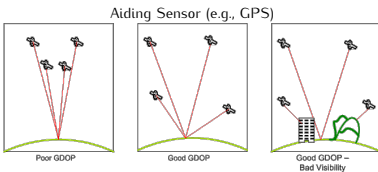
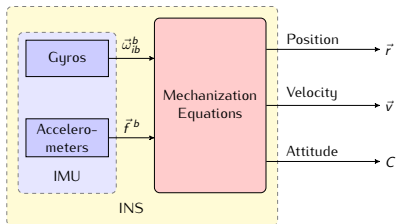
$$\delta \dot{\vec{x}}_g = F_{gg} \delta \vec{x}_g + \vec{w}_g \quad (15)$$

The matrices F_{aa} and F_{gg} are specific to accelerometers and the gyroscopes and their specific configuration within the IMU.

After state augmentation

$$\begin{pmatrix} \delta \dot{\psi}_{eb}^e \\ \delta \dot{v}_{eb}^e \\ \delta \dot{r}_{eb}^e \\ \delta \dot{x}_a \\ \delta \dot{x}_g \end{pmatrix} = \begin{bmatrix} -\Omega_{ie}^e & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & -\hat{C}_b^e F_{\psi g} \\ -[\hat{C}_b^e \hat{f}_{ib}^b \times] & -2\Omega_{ie}^e & \frac{2g_0(\hat{L}_b)}{r_{eS}^e(\hat{L}_b)} \frac{\hat{r}_{eb}^e}{|\hat{r}_{eb}^e|^2} (\hat{r}_{eb}^e)^T & -\hat{C}_b^e F_{va} & 0_{3 \times 3} \\ 0_{3 \times 3} & \mathcal{I}_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} \\ 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & F_{aa} & 0_{3 \times 3} \\ 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & F_{gg} \end{bmatrix} \begin{pmatrix} \delta \vec{\psi}_{eb}^e \\ \delta \vec{v}_{eb}^e \\ \delta \vec{r}_{eb}^e \\ \delta \vec{x}_a \\ \delta \vec{x}_g \end{pmatrix} + \\
 \begin{bmatrix} -\hat{C}_b^e & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} \\ 0_{3 \times 3} & -\hat{C}_b^e & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} \\ 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} \\ 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & \mathcal{I}_{3 \times 3} & 0_{3 \times 3} \\ 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & \mathcal{I}_{3 \times 3} \end{bmatrix} \begin{pmatrix} \vec{\zeta}_g \\ \vec{\zeta}_a \\ 0_{3 \times 1} \\ \vec{w}_a \\ \vec{w}_g \end{pmatrix} \\
 = F(t)\vec{x} + G\vec{w}$$

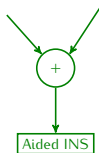
(16)

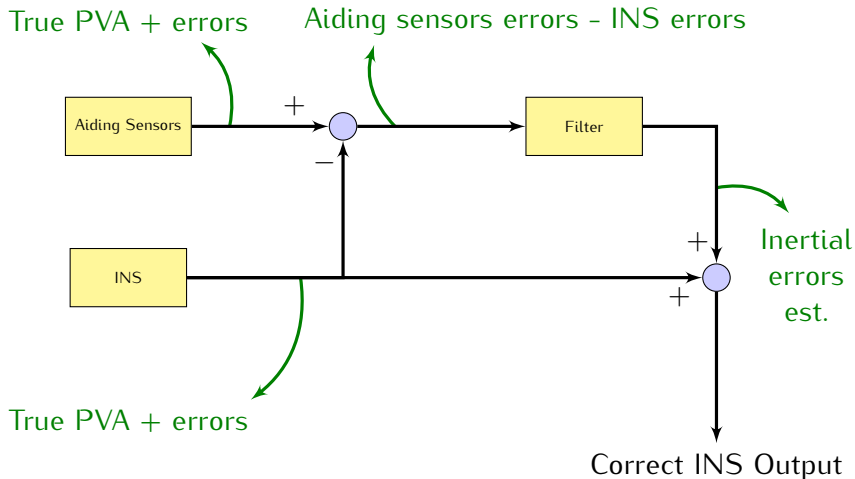


http://nptel.ac.in/courses/105104100/lectureB_11/B_11_3GDOP.htm

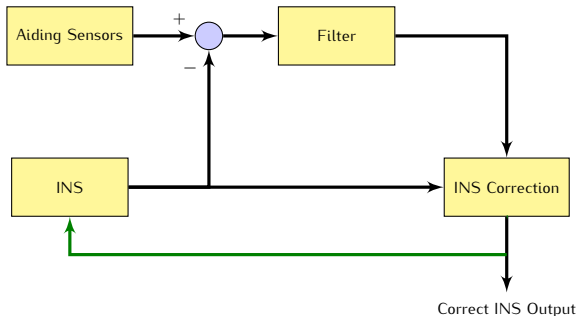
Advantages	Disadvantages
Immune to RF Jamming	Drifts
High data rate	Errors are time dependent
High accuracy in short term	Need Initialization

Advantages	Disadvantages
Errors time-indep.	Sensitive to RF Interference
No initialization	No attitude information





If error estimates are fed back to correct the INS mechanization, a reset of the state estimates becomes necessary.



Project Ahead

$$\hat{\mathbf{x}}_{k|k-1} = \mathbf{\Phi}_{k-1} \hat{\mathbf{x}}_{k-1|k-1} \quad (17)$$

$$\mathbf{P}_{k|k-1} = \mathbf{Q}_{k-1} + \mathbf{\Phi}_{k-1} \mathbf{P}_{k-1|k-1} \mathbf{\Phi}_{k-1}^T \quad (18)$$

Update

$$\hat{\mathbf{x}}_{k|k} = \hat{\mathbf{x}}_{k|k-1} + \mathbf{K}_k (\mathbf{z}_k - \mathbf{H}_k \hat{\mathbf{x}}_{k|k-1}) \quad (19)$$

$$\mathbf{P}_{k|k} = (\mathbf{I} - \mathbf{K}_k \mathbf{H}_k) \mathbf{P}_{k|k-1} (\mathbf{I} - \mathbf{K}_k \mathbf{H}_k)^T + \mathbf{K}_k \mathbf{R}_k \mathbf{K}_k^T \quad (20)$$

$$\mathbf{K}_k = \mathbf{P}_{k|k-1} \mathbf{H}_k^T (\mathbf{H}_k \mathbf{P}_{k|k-1} \mathbf{H}_k^T + \mathbf{R}_k)^{-1} \quad (21)$$

Since the errors are being fed back to correct the INS, the state estimate must be reset after each INS correction.

$$\hat{\vec{x}}_{k|k-1} = 0 \quad (22)$$

$$P_{k|k-1} = Q_{k-1} + \Phi_{k-1} P_{k-1|k-1} \Phi_{k-1}^T \quad (23)$$

$$\hat{\vec{x}}_{k|k} = K_k \vec{z}_k \quad (24)$$

$$P_{k|k} = (I - K_k H_k) P_{k|k-1} (I - K_k H_k)^T + K_k R_k K_k^T \quad (25)$$

$$K_k = P_{k|k-1} H_k^T (H_k P_{k|k-1} H_k^T + R_k)^{-1} \quad (26)$$

$$\Phi_{k-1} \approx I + F\Delta t \quad (27)$$

$$Q = \begin{pmatrix} n_{rg}^2 I_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} \\ 0_{3 \times 3} & n_{ag}^2 I_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} \\ 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} \\ 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & n_{bad}^2 I_{3 \times 3} & 0_{3 \times 3} \\ 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & 0_{3 \times 3} & n_{bgd}^2 I_{3 \times 3} \end{pmatrix} \quad (28)$$

where Δt is the sample time, n_{rg}^2 , n_{ag}^2 , n_{bad}^2 , n_{bgd}^2 are the PSD of the gyro and accel random noise, and accel and gyro bias variation, respectively.

Assuming white noise, small time step, G is constant over the integration period, and the trapezoidal integration

$$Q_{k-1} \approx \frac{1}{2} \left[\Phi_{k-1} G_{k-1} Q(t_{k-1}) G_{k-1}^T \Phi_{k-1}^T + G_{k-1} Q(t_{k-1}) G_{k-1}^T \right] \Delta t \quad (29)$$